

## **Adaptive Control**

Åström, Karl Johan

1979

Document Version: Publisher's PDF, also known as Version of record

Link to publication

Citation for published version (APA): Aström, K. J. (1979). Adaptive Control. (Technical Reports TFRT-7183). Department of Automatic Control, Lund Institute of Technology (LTH).

Total number of authors:

General rights

Unless other specific re-use rights are stated the following general rights apply:

Copyright and moral rights for the publications made accessible in the public portal are retained by the authors and/or other copyright owners and it is a condition of accessing publications that users recognise and abide by the legal requirements associated with these rights.

• Users may download and print one copy of any publication from the public portal for the purpose of private study

- You may not further distribute the material or use it for any profit-making activity or commercial gain
   You may freely distribute the URL identifying the publication in the public portal

Read more about Creative commons licenses: https://creativecommons.org/licenses/

Take down policy

If you believe that this document breaches copyright please contact us providing details, and we will remove access to the work immediately and investigate your claim.

ADAPTIVE CONTROL

K, J ASTRÖM

DEPARTMENT OF AUTOMATIC CONTROL LUND INSTITUTE OF TECHNOLOGY DECEMBER 1979 ADAPTIVE CONTROL

K.J. Aström

Distribution by (name and address)

## ADAPTIVE\_CONTROL

the changing epresent in Œ. changes nonlinear can to t adapt **en**vironment 0 90 to dynamics type parameters disturbances special <u>.</u>2 process its changes Œ. the can alter IJĩ. the ·--40 The control characteristics ij system which environment. variations Adaptive

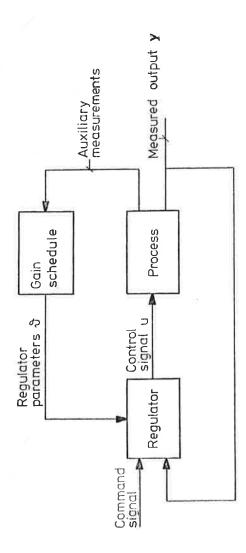
9 and ᆵ 40 constant operates moderate linear feedback dynamics Control changes Mach-number 40 constant which handle with presence introducing satisfactorily. the example. The system aircraft system can situations where ŋ with altitudes. The that reason for control a D typical drastically dynamics. for large not work control and wellflight S one ø speeds however, many MOrk controller will iù H feedback changes process fact are aircraft ∢ 40 not pressure. dynamics in ranges Will ij airplane ŧń A normal supersonic parameters ations variations are, wide Feedback process dynamic There

to process would: Typical chemical parameters. nseq processes regulators it industrial be WOO F. i E 40 production, fixed changes in heat transfer due activity also ageing and given operating condition most Gan for with the 0 useful catalyst control regulators retune depend to t Adaptive also i. holdup times to due មា មា are variations well with desirable changes control production changes. Slow for delay and and Adaptive Φ controlled Ď compensate In reactions howevers examples control. Since

other Ü W) systems mechanical and alves > Σ  $\cdot \vdash \!\!\!\vdash$ Wear sediments examples

Gain 9 the the ary auxiliary changes ations that ٠, eliminate auxi 0 4 vari changing extensi the . scheduling schedule from Fig find the eliminating with ţ 90 to 9 ģ correlate well incorrect possible possible functions gain seen variations th th 40 þ called considered method an then Can sometimes ú) Ú) for It This which W regulato -104 parameter îÙ correct -1 compensation It ŧń . Бe dynamics stem --┯네 dynamics It the to Fig could S scheduling. 40 way See m 40 i, 0 feedforward process parameters process influences scheduling variables variables S. there Gain. 'n ij

system adaptive useful Ŋ very whether u e Φ 9 in C nomenclature considered neverthel ŲÌ •= in. scheduling should controversy scheduling Σ Gai m not W -14 There ain 0 system Ö with



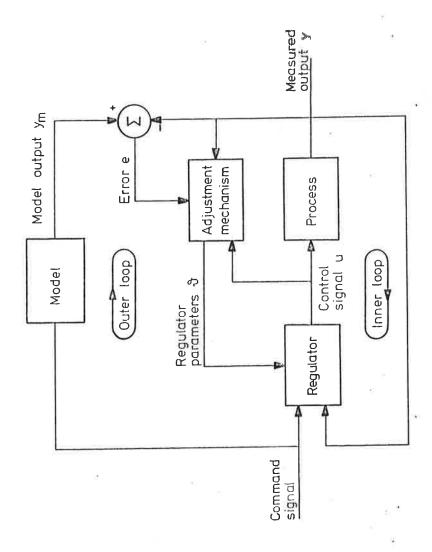
parameter where scheduling. system T) ς a L 40 0 3 Block diagram re eliminated are Figure\_\_1. variations

and Fact in the i) control used and 200  $\subseteq$ d D ī ati and Mach-number flight IJì look Vari -14 sensors I.t parameter the ations table the 9 airdata Vari The parameters Case ģ handle parameter measured by that determined t t In systems. method eliminate variables. are then pressure predominant control interpolation 40 are scheduling technique dynamic flight system

Ned 自ての the ill H ent the ill Fred þ interpolation be possible control also must scheduling normali coeffici obtain variable 40 measurements The controllers must sometimes ed ٠ t o appl constant control using rements. system gain The measurements .H Ü auxiliary ÚÌ. condition. systems based on ·H ģ The measn the It linear # normalized consuming. before scheduling operation of simulations. normalized The process Φ operating 40 parameters. retransformed The output time gain with the drawback with variables. the extensive safe Ü Ü each the Ď the the design dimension-free for W) together driven iù H þ obtain and ated normalized one the fied ũ designed abl process system method Cul nsed that vari vari to

al TŲ are the ordinary model control signal 90 NO K specifications In thought command the ψì e P tel MRAS 40 Ŋ □. which the can be part loop SYStems. to t dynamic model W inner should respond regulator model The the reference adaptive stems reference loops. The ideally Š σ Ń reference ή Σ 40 adaptive Fig. the terms 40 output See that consisting Model reference in process given Notice system

From The adjustments the S ·H Show between pecomes loop. mechanism zero regulator. t t drive Feedback regulator error 40 easy > to output The adjustment error linear ĵĝ. the Ŵ ŋ U) attempt adjusted and 1 i ke It the that process simple process parameters nontrivial looks which May the brings ator can be determine the ø also Ŋ with the such loop and which controller thus ∭ •#4 solved 40 c problem regul outer ·H system Ξ composed loop to  $\rightarrow$ parameters þ W the output outer the the problem not This stable 40 i D loop t a Can The parameters model regulator error made obtained key ŋ ij control . small that that The are the the



system adaptive reference mode1 diagram of Block . Eigure\_2. (MRAS).

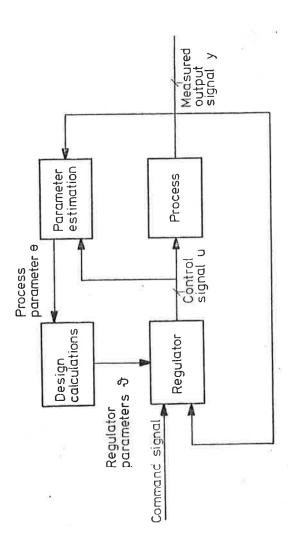
6 ë adjustment mechanism; MRAS: original the parameter in nsed following Ų) m Z • 'MIT-rule' The

$$ds = -k \frac{3}{a^{3}} \cdot e, \quad i = 1, ..., n.$$

many regulator which represents three sensitivity multiplier, for are composed of parameter cal The Equation (1) adjustable the m e/93 typi and outputs, ŋ computing mechanism which is .H errors iù M configuration the × the inputs are adaptation rate. for derivatives and in E = 2 process 96. linear filter integrator. This adjustment À >> || From systems. the Œ variables derivatives parameter sensitivity parameters, determines parts: a adaptive an and The

٥ closed (1979) stability Lyapunov unstable recently 40 using theory 92 is, however, only give modified the in unfortunately made can be been It rule stability theory. real progress has Dan systems. The The MIT-rule system. adaptive Popov loop that for

the two an composed self-tuning calculation. 90 and οf composed The parameters IJì •≓ process which design The loop; the iù iù ŋ 90 and 40 regulator. outer STR thought estimator consists the Self-tuning\_\_regulators. be adjusted by feedback loop rameter Can 3, inner œ C ordinary linear regulator, Fig. recursive are loops. The regulator



(STR) egulator ۲ -tuning ü 56] T) 4 Ö agram <u>0</u> Block ומו Figure

are the 50 Ö estimation estimator any considerably model • eliminated such regulat regulat έ It There for are Ą scheme **6**] ⊆ 40 Bog mechanism here 0 the ۲ parameter T) process are Ø M then imation 9 process simil 4 based D call 90 calculations i. because and very W algorithm est the terms -the ÚÌ 44 justment 101 regulator ---N ú 4 M reparametrize exel . Ш ----Fig because self-tuners paramet ij O Fi It U ğ ğ design Φ. E E Compa  $\Box$ self-tuning expressed model e e • called stem 9 and shown Uì the T dentification process 40 the design îù > -Ŵ nterpreted U1 because ·H 5\_ adaptive paramete regulator Ù 90 **b**1 Ú) -tuner ·H 40 The possi ij) 4 variant ons implic sel ified -parameters ator nati sometimes that erence The 0 Œ Œ, ā ā regul simpl combi based Such many can May 40

ept ordinary era × Sev are  $\subseteq$ 9 -14 Ų estimat nseq becomes be parameter can ator regul regulators the the 4 -14 Since feedback f-tuning . Mays gain sel ifferent The constant O

constant With controller different run until EÚ also Vary with Case self-tuner is algorithms also be H. O) with run iù M purpo tuning course this the ij The for ij Gan then systems process which with loop. In whose control 40 points. ΣœM the settings schedule. The system is self-tuner obtained. The Can this left for Since tuner self-tuner possible to use In controller operating control system is II) stored. regulator The the parameters obtained. ŲΝ Ū Ω Η· ta Œ. parameters. nseq 40 The performance connected different 自に回 adaptive the parameters suitable conditions. (A) -≓ Ď obtained gain and can įţ adjustable at rue automatically, iji H O satisfactory Ų disconnected ם a t -14 the self-tuner self-tuner parameters obtain parameters regulator operating constant, build ψ ψ adjust many to t to

The 9 its unified done using the expected value would states and structures model system It m, principle, be stochastic function of From arguments. The Regulator regulators to minimize theory. Œ. heuristic then described by framework. This can, in scalar adaptive control control the m Ö ů) O il I criterion is formulated based which obtain stochastic are function are Stochastic to STR envi ronment theoretical nonlinear appealing and controls a loss

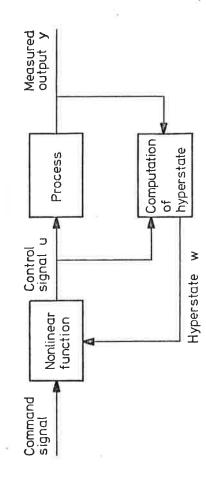
the minimizes expl Under general. Useful control which not known in difficult. rij 13 are finding function existence 40 The problem for 1055 conditions expected

for dynamic Bellman 90 simple Feedback conditional maps solve the The the optimal necessary to measurements, problem. 40 because ď space quantity equation updating which ٩ 2 thought equation over obtained very the the the state using variables. Œ, distribution function ŋ interest iп called functional the 90 iğ. can be and the ·H W for 40 where the anly derived ijţ hyperstate which Bellman from the formula 1. N structure estimator generates contral 40 solution W. nonlinear control problem controller state solved numerically ģ nevertheless M) the hyperstate, Ŋ which the problem example Ŵ 90 the ueu exist the E O the i.e. space the called estimator Œ equations solve The 90 function .H. solution for Ą parts, of into estimator, distribution For ig ig 4. the stochastic regulator ij ij to simplicity ii. See Fig. introducing into high dimension. approach the hyperstate W insight þ t wo 1055 Thi T) ribution The and that n a n hyperstate the 40 Feedback optimal programming. nonlinear probability hyperstate, ssumption The controller structural regulator. determine ت 12 disti equation, 90 composed gives prize caseo very This the the The œ

∢ equation common interest some Bellman 2 Û **a** considerabl illustrat described the solve 40 process 40 to pasn are fficult Đ Đ Consider solutions Will di iğ. approximations. example it pproximative Since imple

$$y(t+1) = y(t) + bu(t) + e(t),$$
 (2)

be noise, Can white 8 Q) Equation output, parameter. the > control, onstant the 10 in M ·H J Ф where and



stochastic dynamical s a static a function ŲÌ iù M nonlinear W) --The regulator rol variable as signal. From generated optimal control inputs. the contr command . ນ 9 40 hyperstate the d y as gives diagram and u and which g hyperstate The Block <u>Eigure 4.</u> Blocontroller. system using nonlinearity the

with integrator Ø. minimi ů D 90 to mode1 þ criterion data sampled the Let T) i) i) gain. interpreted unknown

$$\lim_{N \to \infty} \mathbf{E} \frac{1}{N} \sum_{1} \mathbf{y}^{2}(\mathbf{t}). \tag{3}$$

which l av control the Known in. Д à given parameter ij 6 the Ü S ΙŁ minimi

$$u(t) = -\frac{1}{b} y(t)_{\#}$$
 (4)

given can In ijţ mean salved distribution ٥ with  $\sim$ rob1 ,P(t) b<sub>3</sub> be ā. , b(t) gaussian ٥f the Can 4 stribution (y(t) equation prior 0 hyperstate W ·m gaussian triple ţ di Bellman time conditional the The πį to has ь P(t) the zed п Ω outputs characteri the Case variance parameter that simple and and D e follows the inputs then this â(t) **4** I

only the the the the be inject give 90 and When not not W) also uncertainty however, called <u>dual\_control</u> errors Will can willzero. wi11 optimal control law obtained control control signal ta regulator has, the close way. It to reduce the law This is output between keeping are uncertain the The simple control estimation errors small. the system parameter estimates. The the property. bring The in right balance characterized signals into parameters 40 numerically interesting attempt

The following control law

$$u(t) = -\frac{1}{-1} - y(t)$$
 (5)

the <u>ል</u> parameters estimat simply called interpreted case of known and substituting the known parameters with their in in obtained This control Û. can be the It solving the control problem in approximative solution. \_\_control self-tuning regulator certainty-equivalence control certaintyrequivalence ٥ The

The control law

$$u(t) = -\frac{\hat{b}(t)}{-2} - y(t) = -\frac{1}{2} - \frac{\hat{b}^2(t)}{-2} - y(t).$$
 (6) 
$$\hat{b}^2(t) + P(t)$$

cautious\_control law minimizes estimates and uses lower gain when the cautious control another approximation, which is called that the because it hedges Notice criterion uncertain. the

$$E[y^2(t)|y(t-1), y(t-2), ...]$$

(2)

but that it is not optimal for (4).

40 the ij precise nevertheless availability control implement force field is ways and adaptive <del>نا</del> driving The it possible meaning is subject to controversy. The a strong Word in many different development. the\_art. The .H which make economically. rapid nsed 90 microprocessors, State\_\_of\_ unfortunately state controllers ф ij

ů Z infancy. conceptual research towards its appropriate ij problem. Much taken still theoretical problems. in in been the control the stability recently develop adaptive and the key to have 90 understanding of however, needed The theory steps Framework Major

controllers experimental industrial announced. MRAS, adaptive control based on adaptive techniques on pilot plants, A few adaptive have also recently been a reasonable number of aerospace systems: based on microprocessors feasibility studies of been There have and other processes, and

## References

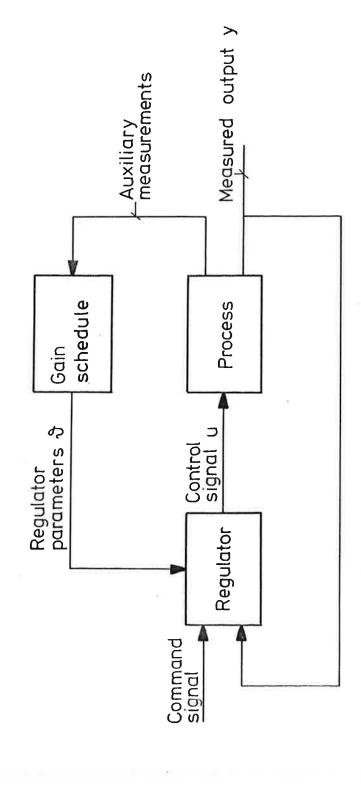
- 8. Aström. K.J., Borisson, U., Ljung, L., and Wittenmark, I Theory and applications of self-tuning regulators. <u>Automatica</u>, 13 (1977) 457-476.
- Springer Egardt, B., Stability\_of\_Adaptive\_Controllers. Verlag, Berlin, 1979.
- Landau: I.D.: Adaptive\_Control\_\_\_The\_Model\_Reference Approach. Marcel Dekker Inc.: 1979.
- Narendra, K.S. and Monopoli, R.V. (editors), <u>Applications\_of</u> <u>Adaptive</u> Control, Academic Press, 1980.

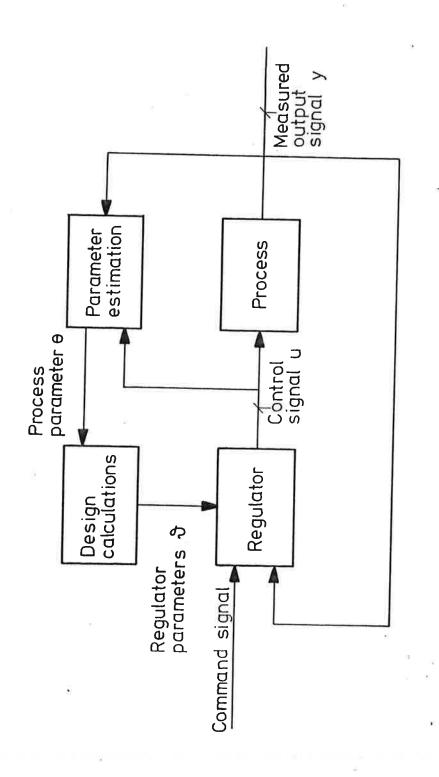
## Adaptive\_control

the characteristics alter its parameters to adapt to a changing environment in environment can represent variations system which nonlinear control in dynamics or changes disturbances. type of The changes process A special 9

drastically with control spaads feedback. There are, however, many situations where the process dynamics are so large that a constant moderate introducing example. linear feedback controller will not work satisfactorily Well of 90 typical presence flight system with constant parameters will not work control system can handle ranges in fact one reason for Œ wide σ dynamics of the airplane changes in process dynamics. The supersonic aircraft is Mach-number and dynamic pressure. over which operates A normal feedback variations is altitudes Control of a aircraft changes in variations and

due to ageing to retune regulators when production changes. Adaptive control most processes depend on industrial process controlled well with regulators with ij would; however; be desirable holdup times are variations also be used to compensate for changes In a given operating condition control is also useful for delay and Typical examples Since production, it parameters WOBT. control. Adaptive be and





(3)